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(71) Applicant: **CORDANT TECHNOLOGIES INC.**
[US/US]; Gateway Towers West, Suite 1600, 15 W. South Temple, Salt Lake City, UT 84101 (US).

(72) Inventor: **GUILLOT, David, G.**; 101 Amber Avenue, Tremonton, UT 84337 (US).

(74) Agents: **TAYLOR, David, Scott et al.**; Sullivan Law Group, Suite 120, 5060 North 40th Street, Phoenix, AZ 85018 (US).



A1

(54) Title: **METHOD OF INSULATING A CASE OF A SOLID PROPELLANT ROCKET MOTOR**

(57) Abstract: This method permits manufacturing EPDM rocket motor insulation in which carbon fibers are dispersed and immobilized in the EPDM polymeric matrix, but are not excessively fractured or fragmentized, i.e., broken into smaller fragments, when encountering degrees of shear necessary to homogeneously or otherwise distribute or disperse the carbon fibers in the EPDM polymeric matrix. The method is substantially solvent free, and is performed via distributive/reduced shear mixing to distribute the fragile carbon fibers into a rubber matrix without excessive damage. According to one embodiment, at least about 50 % of the elastomer composition introduced into the mixing apparatus is liquid EPDM terpolymer having sufficiently low molecular weight and high diene content to permit dispersion of the carbon fibers in the EPDM without substantial fragmentation of the fibers. According to another embodiment, mixing takes place in a kneader capable of rotating a screw having a discontinuous screw-thread about the screw axis while superimposing an axial reciprocating stroke to the screw. The kneader imparts low shear distributive mixing of the carbon fibers in the EPDM terpolymer.

WO 01/46279

METHOD OF INSULATING A CASE OF A SOLID PROPELLANT ROCKET MOTOR

RELATED APPLICATION

5 Priority is claimed of U.S. provisional application 60/171,619 filed in the U.S. Patent & Trademark Office on December 23, 1999, the complete disclosure of which is incorporated herein by reference.

ORIGIN OF THE INVENTION

The U.S. Government has a paid-up license in this invention and the right in
10 limited circumstances to require the patent owner to license others on reasonable terms as provided by the terms of F04611-97-C-0053 to the Air Force Rocket Laboratory.

BACKGROUND OF THE INVENTION

1. Field of the Invention

This invention is directed to a process for making elastomer-based insulation for rocket motors, and in particular to a process in which fragile carbon fibers are mixed with and preferably homogeneously dispersed in EPDM without requiring the use of a volatile solvent for dissolution of EPDM during fiber incorporation. The insulation of this invention is especially useful for placement in the nozzle or case,
20 including between a solid propellant grain and a rocket motor case for protecting the case from high temperatures experienced during burning of solid propellant grains.

2. Description of the Related Art

Solid rocket motors typically include an outer case or shell housing a solid propellant grain. The rocket motor case is conventionally manufactured from a rigid, yet durable, material such as steel or filament-wound composite. The propellant is housed within the case and is formulated from a composition designed to undergo combustion and thereby produce the requisite thrust for attaining rocket motor propulsion.

During operation, a heat insulating layer (insulation) protects the rocket motor case from heat and particles streams generated by the combusting propellant.
30

Typically, the insulation is bonded to the inner surface of the case and is generally fabricated from a composition capable of withstanding the high temperature gases produced when the propellant grain burns. A liner layer (liner) functions to bond the propellant grain to the insulating layer and to any non-insulated portions of the case,
5 as well as to inhibit interfacial burning. Liner compositions are generally known to those skilled in the art. An exemplary liner composition and process of applying the same to a case is disclosed in U.S. Patent No. 5,767,221, the complete disclosure of which is incorporated herein by reference to the extent that it is compatible with this specification.

10 The combustion of solid rocket propellant generates extreme conditions within the case of the rocket motor. For example, temperatures inside the rocket motor case typically reach 2760°C (5,000°F), and interior pressures may exceed 1,500 psi. These factors combine to create a high degree of turbulence within the rocket motor case. In addition, particles are typically entrained in the gases produced during the propellant
15 combustion. Under the turbulent environment, these entrained particles can erode the rocket motor insulation. If the insulating layer and liner are pierced during rocket motor operation, the casing is susceptible to melting or degradation, which can result in failure of the rocket motor. Thus, it is crucial that insulation compositions withstand the extreme conditions experienced during propellant combustion and
20 protect the case from the burning propellant. It is also crucial that insulation compositions possess acceptable shelf life characteristics such that it remains sufficiently pliable, without becoming fully cured, until used in application to the rocket motor casing. This requirement is essential because the production of a given lot of insulation may have to wait in storage for a number of months prior to use.
25 Typically, the insulation may be stored in large rolls in an uncured, or at most a partially cured, state until ready for use. A number of curing agents are well known and are conventionally employed but still must be compatible with the overall EPDM formulation to permit satisfactory shelf life. This in turn requires a balancing of curing agent activity.

30 In the past, attempts at producing insulating materials that would protect the rocket motor case focused on filled and unfilled rubbers and plastics such as, phenolic resins, epoxy resins, high temperature melamine-formaldehyde coatings, ceramics,

polyester resins, and the like. These plastics, however, crack and/or blister as a result of the rapid temperature and pressure fluctuations experienced during combustion.

Elastomeric compositions have also been used as rocket motor insulation materials in a large number of rocket motors. The elastomeric compositions have
5 been selected because their mechanical, thermal, and ablative properties are particularly suited for rocket motor applications. However, the ablative properties of elastomers are often inadequate for rocket motor operation. For example, insulation, whether thermosetting or thermoplastic, are characterized by relatively high erosion rates unless reinforced with a suitable filler. The criticality of avoiding such high
10 erosion rates is demonstrated by the severity and magnitude of the risk of failure due to erosion. Most insulation is of necessity "man-rated" in the sense that a catastrophic failure can result in the loss of human life -- whether the rocket motor is used as a booster for launch of the space shuttle or is carried tactically underneath the wing of an attack aircraft. The monetary cost of failure in satellite launches is well-publicized
15 and can run into the hundreds of millions of dollars.

In order to improve the ablative properties of elastomeric compositions, it has been proposed to reinforce the elastomeric compositions with fillers, such as organic-based fibers or carbon fibers. For instance, an exemplary carbon fiber-filled rocket motor insulation composed of solid NORDEL 1040 as the primary terpolymer is
20 commonly known in the industry as the STW4-2868 thermal insulation and has the following composition:

TABLE 1
STW4-2868 THERMAL INSULATION FORMULATION
(carbon fiber; parts by weight)

Ingredient	Function	Parts by Weight
NORDEL 1040	Primary EPDM terpolymer base	80
Neoprene FB	Secondary polymer base	20
Zinc oxide	Activator	5
Sulfur	Curative	1
HAF carbon black	Pigment	1
MBT	Accelerator	1
AGERITE Resin D	Antioxidant	2
AGERITE HPS	Antioxidant	1
Tellurac	Accelerator	0.50
Sulfads	Accelerator	0.75
VCM carbon fibers	Filler	41
Total Parts by Weight		153.25

5 Although organic-based fibers can be dispersed within the EPDM without too much difficulty, the homogeneous dispersion of carbon fibers in an elastomeric composition presents a difficult processing problem. The mixing process is complicated by the fragility of the carbon fibers. Mixing of carbon fibers into a solid elastomer under high shear physically deteriorates the carbon fibers into smaller
10 particles or shreds, thereby negating the advantageous physical attributes that the carbon fibers would otherwise have contributed to the insulation.

Conventionally, the problem of carbon fiber fragility has been addressed by dissolving the elastomer into solution with an appropriate organic solvent to lower the viscosity of the elastomer or elastomer mixture. Suitable solvents include, by way of
15 example, hydrocarbons such as hexanes, heptanes, and/or cyclohexane. The frangible graphitized carbon fibers can then be mixed with the solution in, for example, a sigma-blade mixer without significant breakage of or damage to the carbon fibers. The material is then sheeted out and the solvent allowed to evaporate at ambient atmosphere or in an oven.

20 The use of solvent in this processing technique presents several drawbacks. For example, solvent processing techniques, such as those conventionally used to

disperse carbon fibers in EPDM (ethylene propylene diene monomer) rubber, are relatively expensive. Material costs are increased by the use of solvents, as are processing costs, since additional workers and equipment are required to handle and process the solvents. Further, considerable costs and worker safety issues are 5 associated with the disposal of hazardous volatile organic solvents.

Thus, although it has been long recognized that carbon fiber-filled EPDM is an excellent candidate for such rocket motor insulation, a need persists in the art for a low cost and non-hazardous solvent-free synthesis route that produces EPDM insulation having carbon fibers homogeneously dispersed therein, but without being 10 subject to significant breakage or damage.

SUMMARY OF THE INVENTION

It is, therefore, an object of this invention to fulfill the long-felt need in the art outlined above by providing a method of manufacturing EPDM rocket motor insulation in which carbon fibers are dispersed and immobilized in the EPDM 15 polymeric matrix, but are not excessively fractured or fragmentized, i.e., broken into smaller fragments, when encountering degrees of shear necessary to homogeneously or otherwise distribute or disperse the carbon fibers in the EPDM polymeric matrix.

The above and other objects are attained by a substantially solvent-free process in which the insulation is manufactured via distributive/reduced shear mixing 20 to distribute the fragile carbon fibers into a rubber matrix without excessive damage.

In accordance with one embodiment of this substantially solvent-free process, the elastomer composition comprises carbon fibers and EPDM terpolymer, at least 50 wt% of which is introduced as an ingredient into the mixing apparatus as liquid EPDM terpolymer having sufficiently low molecular weight and high diene content to 25 permit dispersion of the carbon fibers in the EPDM without substantial fragmentation of the fibers. As referred to herein, liquid EPDM means EPDM terpolymer that is flowable at room temperature. Suitable mixing apparatuses for this embodiment include sigma-blade and vertical-blade mixers. Certain kneaders, such as discussed below in connection with the second embodiment, capable of superimposing a 30 rotational and axial mixing motion to the carbon fibers can also be used.

In accordance with another embodiment of the invention, the elastomer composition is prepared, optionally under substantially solvent-free conditions with little or no liquid EPDM terpolymer, by use of a kneader capable of rotating a screw having a discontinuous screw-thread about the screw axis while superimposing an axial reciprocating stroke to the screw. This kneader imparts low shear distributive mixing of the carbon fibers in the EPDM terpolymer. The kneader used in this embodiment is especially suitable where little or no liquid EPDM ingredient and no volatile solvent are included in the formulation.

As referred to herein, carbon fibers are fibers having been subject to at least substantial graphitization or carbonization, and preferably have about 98 wt% or more carbon content.

Other objects, aspects and advantages of the invention will be more apparent to those skilled in the art upon reading the detailed description and appended claims which, when read in conjunction with the accompanying drawings, explain the principles of this invention.

BRIEF DESCRIPTION OF THE DRAWINGS

The accompanying drawings serve to elucidate the principles of this invention.

In such drawings:

FIG. 1 is a schematic cross-sectional view of a rocket motor assembly in which the insulation is provided;

FIG. 2 is a schematic cross-sectional view of a kneading apparatus suited for use with this invention;

FIG. 3 is a cross-sectional view taken along sectional line III-III in FIG. 2;

FIG. 4 is a schematic view of an axial segment of a discontinuous screw barrel of the kneading apparatus of FIG. 2, with the axial segment section being projected onto a flat plane for explanatory purposes;

FIG. 5 is the schematic view of FIG. 4, with kneading pins of the kneading apparatus being superimposed onto the illustrated axial segment;

FIG. 6 is the schematic view of FIG. 5, showing the paths of relative movement of selected ones of the kneading pins relative to the discontinuous screw barrel, and in particular relative to the screw flights of the barrel; and

5 FIG. 7 is a schematic cross-sectional view of a char motor used in testing examples discussed hereinbelow.

DETAILED DESCRIPTION OF THE INVENTION

The present insulation compositions 10, when in a cured state, are especially suited for disposal on the interior surface of the rocket motor case 12, as shown in FIG. 1. Typically, a liner 14 is interposed between the insulation 10 and propellant 10 16. The insulation 10 and the liner 14 serve to protect the case 12 from the extreme conditions produced by the propellant 16 as it undergoes combustion reactions and is exhausted through nozzle assembly 20. Methods for loading a rocket motor case 12 with the insulation 10, the liner 14, and the propellant 16 are known to those skilled in the art, and can be readily adapted within the skill of the art without undue 15 experimentation to incorporate the insulation composition of this invention.

Unlike conventional techniques that make use of solvent within a mixing apparatus to achieve adequate distribution of carbon fibers in solid EPDM ingredients without significant fiber fragmentation, it is an object of this invention to achieve distribution of carbon fibers in an EPDM matrix under solvent-free conditions, or at 20 least substantially solvent-free conditions. As referred to herein, substantially solvent-free means that the process is performed with a sufficiently small amount of volatile solvent that, even if the volatile solvent is not removed during manufacture of the insulation, the volatile solvent will not be present in a sufficient amount to violate applicable environmental or safety regulations during manufacture, rocket motor 25 storage, or rocket motor operation due to volatilizing of the solvent. Generally, the term substantially solvent-free preferably means not more than about 5 wt% of volatile solvent based on the dry ingredients of the insulation. Preferably, the process is conducted completely free of volatile solvent.

In accordance with a first embodiment of this invention, this and other objects 30 are achieved by using liquid EPDM as a significant portion of the EPDM ingredients

introduced into the mixing apparatus. The amount of liquid EPDM ingredient used to ensure adequate distribution of the fibers, without accompanying excessive fragmentation of the fibers, depends upon the mixing apparatus used. Generally, where a conventional mixer known in the insulation industry, such as a sigma-blade 5 mixer, is used to disperse the carbon fibers within the EPDM matrix, the insulation composition preferably contains at least about 50% by weight, and more preferably at least about 90% by weight liquid EPDM as an ingredient, based on the total weight of the EPDM (i.e., both the solid and liquid EPDM ingredients). Where a vertical blade mixer is used to disperse the carbon fibers within the EPDM matrix, the insulation 10 composition preferably contains slightly more liquid EPDM, such as at least about 90% by weight, and more preferably at least about 95% by weight liquid EPDM as an ingredient, based on the total weight of the EPDM (i.e., both the solid and liquid EPDM ingredients). Where a kneader such as the one illustrated in FIGS. 2-6 is used, even less of the liquid EPDM (or even no liquid EPDM, as detailed in the second 15 embodiment below) is required to obtain homogeneous dispersion of the fibers without excessive fragmentation, *i.e.*, all of the EPDM can be in a solid state when introduced into the kneader.

Generally, the EPDM, *i.e.*, both the solid and liquid ingredients, comprises from about 35 wt% to about 90 wt%, and still more preferably from about 45 wt% to 20 about 75 wt% of the total weight of the rocket motor insulation. The EPDM terpolymer can be formed from 1,4-hexadiene, dicyclopentadiene, and/or an alkylidene norbornene, such as ethylidene norbornene (ENB), as the diene component. Suitable commercially available liquid EPDM terpolymers are TRILENE 67A and TRILENE 77, available through Uniroyal Chemical. It is noted, however, 25 that a portion or all of the liquid EPDM can be substituted for another liquid polymer ingredient, such as liquid polyurethanes, so long as the substituted liquid polymer ingredient obtains the same distributive function with regard to the carbon fibers without excessive fragmentation. Suitable solid EPDM terpolymers having a 1,4-hexadiene component for use in this invention include NORDEL 1040, NORDEL 30 2522, and NORDEL 2722E, made by DuPont. Suitable solid EPDM terpolymers having an ENB diene component for use in this invention include, without limitation, and as stated above, KELTAN 4506, KELTAN 1446A, KELTAN 2308, each of

which is available from DSM of the Netherlands, and NORDEL IP 4520 and NORDEL IP 4640, both of which are and continue to be available from DuPont Dow Elastomers.

The curing package can include sulfur curing agents and/or peroxide curing agents for crosslinking and/or chain extending polymers or polymer precursors (e.g., prepolymers). A suitable insoluble sulfur curing agent is AKROSPERSE IS-70 from Akro Chem, and CRYSTEX OT-20 available through Charles H. Haynes, Inc. Other forms of elemental sulfur can also be used. Suitable peroxide curing agents include dicumyl peroxide, 2,5-dimethyl-2,5-bis-(t-butylperoxy)hexane, 2,5-dimethyl-2,5-bis-(benzoylperoxy)hexane, 2,5-dimethyl-2,5-di(t-butylperoxy)-3-hexane, n-butyl-4,4'-bis-(t-butylperoxyl)valerate, 4,4'-methyl-bis-(cyclohexylamine)carbomate, 1,1-bis-(t-butylperoxy)-3,3,5-trimethylcyclohexane, α,α' -bis-(t-butylperoxy)-diisopropylbenzene, 2,5-dimethyl-2,5-bis-(t-butylperoxy)hexyne-3, and t-butyl perbenzoate. A commercially available peroxide is available under the trade name Di-Cup 40KE, which comprises about 40% dicumyl peroxide on a clay carrier. (The clay carrier is available from Burgess Pigment Company.) Another suitable curing agent (besides sulfur and peroxide curing agents) is bromomethyl alkylated phenolic resin, available as SP-1056 from Schenectady Int'l, Inc. of Schenectady, N.Y.

In typical formulations, the curing agent comprises from about 0.5 phr to about 8 phr, more preferably about 2 phr to about 5 phr. As referred to herein and generally accepted in the art, "phr" means parts by weight per one hundred parts by weight polymer.

The curing package preferably also includes at least one phosphate cure accelerator. In the case of a sulfur curing agent, the accelerator can be, by way of example, RHENOCURE AP-5, RHENOCURE AP-7, RHENOCURE AP-3, RHENOCURE ZADT/G, and RHENOCURE S/G, which are available from Rhein Chemie and Accelerator VS, available from Akro Chem. Additional cure accelerators that may be used in combination with the phosphate cure accelerator include butyl zimate; benzothiazyl disulfide (commercially known as ALTAX); dithiocarbamate-containing blends (such as AKROFORM DELTA P.M. from AkroChem); and sulfides such as dipentmethylenethiuram hexasulfide (such as SULFAD from R.T.

Vanderbilt). While the use of Accelerator VS was initially unacceptable in some formulations because of the foul odor problem it generated, it has also been now found that such formulations can be prepared with no significant odor if about 1.0 phr magnesium oxide is added thereto.

5 Suitable cure activators for the curing package include metal oxides, such as zinc oxide (e.g., TZFD-88p from Rhein Chemie Corp. of Trenton, N.J), magnesium oxide (e.g., Elastomag 170, from Morton Chemical Co.), and stearic acid (including palmitic acid), which is available from Harwick Standard Distribution Corp. of Akron, Ohio.

10 The carbon fibers are fibers having been subject to at least partial graphitization or carbonization, and preferably have about 98 wt% or more carbon content. The carbon fibers should have lengths suitable for distribution in mixing equipment. Generally, the carbon fibers are preferably non-continuous, and not less than about 1/16 of an inch in length and not more than about 6 inches in length,

15 although these ranges are not exhaustive as to the scope of the invention. Carbon fibers are supplied commercially by several companies, including FORTAFIL fibers (e.g., FORTAFIL 140 and FORTAFIL 144) from Akzo Nobel of Knoxville, Tennessee, carbon fibers available from Amoco of Charleston, S. Carolina; and PANEX 33 ($\frac{1}{4}$ " x 8" or $\frac{1}{4}$ " x 15"), supplied by Zoltec of St. Louis, Missouri.

20 Generally, the carbon fibers are present in an amount of from about 2 wt% to about 50 wt%, more preferably from about 10 wt% to about 30 wt%, based on the total weight of the insulation. The amount of carbon fibers will generally vary depending on the presence of other ingredients, such as char forming agents, especially phosphate fire retardants, which supplement the carbon fibers by imparting desired physical properties to the insulation.

25 The carbon fibers can be used alone or in combination with other materials affecting the ablative and mechanical properties of the insulation. By way of example, suitable materials include polybenzoxazole fibers, polybenzimidazole fibers, aramid fibers, iron oxide, milled glass, silica, ceramic clay, and the like. Suitable silica particles include HiSil 233 available from PPG Industries, Inc. of Lake Charles, Louisiana and hydrophobized silica particles available from Cabot Corporation as

CAB-O-SIL® TS-610, CAB-O-SIL® TG-308F, CAB-O-SIL® TG-720, CAB-O-SIL® TS-500, CAB-O-SIL® TS-530, and CAB-O-SIL® TG-810G; Degussa as AEROSIL R972, AEROSIL R974, AEROSIL R812, AEROSIL R812S, AEROSIL R711, AEROSIL R 504, AEROSIL R8200, AEROSIL R805, AEROSIL R816, AEROSIL 5 R711, and AEROSIL R104; and Tulco Inc. as TULLANOX 500.

Suitable additives that may be added as required or desired include one or more of the following, in various combinations: antioxidants, flame retardants, tackifiers, plasticizers, processing aids, carbon black, pigments, and bonding agents.

Representative antioxidants for improving the longevity of the cured elastomer 10 include, by way of example, diphenylamine reacted with acetone, available as BLE®- 25 Liquid from Uniroyal Chemical; a mixture of mono-, di-, and tri-styrenated phenols, available as AGERITE SPAR from B.F. Goodrich Chemical Co. Other suitable antioxidants include polymerized 1,2-dihydro-2,2,4-trimethylquinoline (AGERITE RESIN D) and mixed octylated diphenylamines (AGERITE 15 STATLITES), each of which is available from R.T. Vanderbilt Co.

Fillers that function as flame retardants, or char forming additives, can be used, if desired, in lesser amounts than most other additives, which makes it easier to formulate the insulation with good mechanical properties. Both inorganic and organic flame retardants are expected to be useful in the present invention. Examples of 20 organic flame retardants include: chlorinated hydrocarbon, available as DECHLORANE®, in combination with antimony oxide (optionally with diisodecyl phthalate (DIDP)) or hydrated alumina (such as Hydral 710 aluminum trihydrate); melamine cyanurate; phosphate and phosphate derivatives, available as PHOSCHEK P/30® (ammonium polyphosphate) produced by Monsanto Chemical, which can be 25 used alone or in combination with pentaerythritol; DECHLORANE PLUS 25 from Occidental Chemical Corporation of Niagara Falls, N.Y.; and silicone resin, such as DC4-7051 available through Dow Corning. An example of an inorganic flame retardant is zinc borate, such as FIREBRAKE® ZB from U.S. Borax.

Examples of suitable tackifiers are WINGTACK 95 and AKROCHEM P-133. 30 Other ingredients, such as pigments and extruder processing aids (e.g., ARMEEN 18-D) well known in the art and/or suitable for use in rocket motor thermal insulation

applications and extruder techniques, are intended to be included within the scope of the present invention. A suitable modifying elastomer is chlorosulfonated polyethylene, such as HYPALON-20 available from DuPont Dow Elastomers. Non-volatile plasticizers such as hydrocarbon oil, can also be used.

5 The casting of the inventive insulation into a case and curing of the inventive insulation may be performed in accordance with techniques known in the art. As referred to herein and in the appended claims, the inventive composition can be, inter alia, either applied by casting into a rocket motor case then cured, or cured, optionally cut into appropriate geometry and size, the subsequently applied into the rocket motor
10 case.

Referring now more particularly to FIGS. 2-6, the kneader in accordance with a preferred embodiment of this invention is a Buss Kneader® available through Buss Compounding Systems, AG, a plant engineering group of Georg Fischer Plant Engineering. A representative Buss Kneader® brand kneader is model MDK/E-46.

15 This kneader is commercially available, and is currently believed to have been used in the past in various other industries, including the following: construction; electrical and electronic component parts; automotive parts; chemicals; house appliances; foodstuffs, packaging, and consumer goods. Another similar kneader is available from B&P Process Equipment & Systems.

20 The Buss Kneader® brand kneader has a housing module (or barrel) 20 defining a chamber 22. A plurality of additional modules (not shown) having respective chambers can be united together to provide an extended chamber. The housing module 20 can be equipped with a jacket or internal fluid passages for heating. In order to allow for ease in maintenance and operation, the housing module
25 20 can be a split-barrel arrangement to allow opening of the barrel 20 along its length, thereby facilitating access to the chamber 22.

In the illustrated embodiment, a single rotatable screw 24 is received in the chamber 22. Generally, the screw 24 is from about 30 mm to about 200 mm in diameter, and has a length-to-diameter (L:D) ratio of from about 8:1 to about 20:1,
30 although this invention is not so limited, given the flexibility of uniting a desired number of modules 20.

As shown in FIGS. 2-6, the periphery of the screw 24 has a plurality of screw flights 30. The screw flights 30 each have a rhombic configuration in the illustrate embodiment, although the present invention is not thereby limited in scope. As best shown in FIG. 4, the screw flights 30 are arranged relative to each other to provide a plurality of screw flight columns 32. For each of these screw flight columns 32, the respective screw flights 30 thereof are aligned along the longitudinal axis of the screw 24, yet spaced from each other by an axial distance. In a preferred embodiment, the screw 24 has three screw flight columns 32a, 32b, and 32c. The circumferential centers C of the screw flights 30 of screw flight column 32a are positioned about the circumference at intervals of 120° from the circumferential centers of the screw flights 30 of screw flight columns 32b and 32c. Defined between each of the adjacent columns 32a, 32b, and 32c are gaps 34a, 34b, and 34c. Whereas the screw of a conventional single-screw extruder has a continuous spiral or helical screw face extending along its length, the screw 24 of the illustrated embodiment has a discontinuous screw face, with the spiral or helical path of the screw face being interpreted by the gaps 34.

The housing module 20 has kneading pins (also referred to as kneading teeth) 40, which in the illustrated embodiment have diamond-shaped cross sections. Each of the kneading pins 20 extends from an inner peripheral thereof along a respective radial direction of the housing module 20. As shown in FIG. 4, the kneading pins 20 collectively define three kneading pin columns 42a, 42b, and 42c, each spaced 120° from each other about the circumference of the screw 24 and dimensioned so as to be receivable in the gaps 34. The kneading teeth can be hollow and connected to a supply means for permitting the injection of fluid constituents through the kneading teeth and directly into the melt.

During operation, the screw 24 is rotated about its longitudinal axis while an axial stroke is superimposed on the screw 24 to oscillate the screw 24 back and forth in the axial direction. A gear box (not shown), also available with the Buss Kneader® brand kneader through Georg Fischer Plant Engineering, preferably ensures that each revolution of the screw 24 is accompanied by one full forward and backwards stroke of the screw 24. At the same time, the housing module 20 and kneading pins 40 remain stationary relative to the rotating/oscillating screw 24.

The rotating/oscillating movement of the screw 24 causes the kneading pins 40 to traverse across the faces of respective kneading flights 30, thus generating a shear which cleans the faces of the kneading flights 30 and effects dispersion and distributive mixing. This relative movement between the kneading flights and the 5 kneading pins 40 is explained below in more detail with reference to FIG. 6, which shows selected pins 40a and 40c and their respective paths of movement relative to the screw 24. As shown in FIG. 6, the kneading pins 40 moves across the faces of the kneading flights 30 and across the gaps 34, thereby cleaning the faces of the kneading flights 30 and causing dispersion and distributive mixing to take place.

10 As mentioned above, a Buss Kneader® model MDK/E-46 brand kneader having a 46 mm single screw with a process L/D ratio of 11:1 can be used. This model kneader can be used in combination with a Reliance 40 HP 1750 rpm DC Motor and Flex Pak 3000 controller.

15 Vertical feeds can be provided at different axial locations along the length of the housing module 20. Preferably, the inlet feeders are jacketed vertical screw feeders. Generally, the polymeric ingredients and carbon fiber are introduced into the most upstream feed, fire retardants and other additives are added further downstream (along the axial direction of the housing module 20), and the curing package is introduced at the most downstream feed port. In this manner, insulation composition 20 may be continuously produced. The temperature of the chamber is generally set in the range of from about 66°C (150°F) to 93°C (200°F) during operation.

An advantage of using the kneader of this second embodiment is that the insulation composition discharged from the kneader can be introduced directly into an extruder for extrusion of the EPDM material. A suitable extruder for use with the 25 kneader of this second embodiment is a discharge extruder GS70. The ability to extrude in this embodiment provides improvements over conventional techniques, in which the insulation composition is calendered into sheets then cut.

EXAMPLES

30 The following examples illustrate embodiments which have been made in accordance with the present invention. Also set forth are comparative examples

prepared for comparison purposes. The inventive embodiments are not exhaustive or exclusive, but merely representative of the many types of embodiments which may be prepared according to this invention.

TABLE 1

5

(all units in parts by weight)

Ingredient	EXAMPLE						COMPARATIVE EXAMPLE	
	1	2	3	4	5	6	A	B
Trilene 67 [liquid EPDM]	100	100	100	100	100	100	40	
DSM Keltan [solid EPDM]							50	
NORDEL 1040 [solid EPDM]								80
Neoprene FB [plasticizer]								20
Hypalon 20 [polymer]							10	
Panex 33x8 [1/4" fibers]	26							
Fortafil 144 [carbon fibers]		38.5	45	40.5	30	25.7	26.85	
VCM [carbon fibers]								41
Akrochem P-133 [plasticizer/-tackifier]							5	
Agerite Stalite S [antioxidant]	2	2	2	2	2	2	2	1
Agerite Resin D [antioxidant]								2
HiSil 233 [fire retardant filler/-char forming agent]	5	5	5	5	5	5	3	
FireBrake ZB [fire retardant filler/-char forming agent]		19.5						
Hydral 710 Aluminum Trihydrate [filler]		19.5						
Carbon black [filler]								1
DC4-7051 [fire retardant/char forming agent]	5			8.5	5	5	5	

Ferric oxide [filler/pigment]			1.13					
Antimony oxide (4% DIDP) [fire retardant/filler]			18					
Dechlorane Plus 25 [fire retardant filler/char forming agent]			45					
Pentaerythritol PE 200 [fire retardant filler/char forming agent]				8.5				
Phos-Check P/40 [fire retardant filler/char forming agent]				30				
Melamine cyanurate [fire retardant filler/- char forming agent]	25					25	25	
Zinc oxide [activator]							5	5
Kadox 920C zinc oxide [activator]	5	4	4	4	4	5		
Altax [accelerator]	1.1	1.5	1.5	1.5	1.5	1.2		
Accelerator VS	2.7							
Akroform Delta P.M. [accelerator]		0.3	0.3	0.3	0.3	0.25		
Sulfads [accelerator]		0.82	0.82	0.82	0.82	0.82		0.75
Butyl zimate [accelerator]							0.5	
Rhenocure AP-5 [accelerator]							3.5	
Crystex OT-20 [curative]	1.05	1.22	1.22	1.22	1.22	1.22		
SP-1056 [curative]							1.1	
Captax [accelerator]								1
Tellurac [accelerator]								0.5
Sulfur [curative]								1
Total Parts by Weight	172.8	192.3	223.9	202.3	149.8	171.2	178.95	153.25

Examples 1 and 6:

All solid ingredients, with the exception of the TRILENE 67, were blended in a V-shell blender at ambient temperature over several hours. The TRILENE 67 was

separately introduced into a Brabender mixer equipped with a sigma blade operating at 10 rpm and set to 60°C (140°F). The TRILENE 67 was mixed in the Brabender mixer for a sufficient amount of time to warm the TRILENE 67 to 60°C. Next, the blended material from the V-shell blender was introduced into the Brabender mixer 5 and allowed to mix with the TRILENE 67 until the fibers were uniformly dispersed in the TRILENE 67. The formulation was then dumped from the Brabender mixer to a mill for shaping into sheets before cooling. Each sheet was about 1.27 cm (0.5 inch) in thickness.

Examples 2 through 5

10 All solid ingredients, with the exception of the TRILENE 67 and the carbon fibers, were blended in a V-shell blender at ambient temperature over several hours. The TRILENE 67 was separately introduced into a Brabender mixer equipped with a sigma blade operating at 50 rpm and set to 77°C (170°F). The TRILENE 67 was mixed in the Brabender mixer for a sufficient amount of time to warm the TRILENE 15 67 to 77°C. Next, the blended material from the V-shell blender was introduced into the Brabender mixer and allowed to mix with the TRILENE 67. The speed of the Brabender mixer was then slowed to 20 rpm, and the fibers were introduced into the Brabender mixer and mixed until the fibers were uniformly dispersed in the TRILENE 67. The formulation was then dumped from the Brabender mixer to a mill 20 for shaping into sheets before cooling. Each sheet was about 1.27 cm (0.5 inch) in thickness.

Comparative Examples A and B

Comparative Example A was prepared by mixing all of the ingredients, with the exception of the carbon fiber, in a laboratory mixer. The carbon fiber was 25 incorporated into this mixture in a twin screw extruder (containing counter-rotating screws) by adding the mixed polymeric material and the carbon fibers in a single port of the twin screw extruder. Comparative Example B was made by solvent processing with a hydrocarbon solvent.

TABLE II

	EXAMPLE						Comparative Example	
	1	2	3	4	5	6	A	B
Average Ablation Rate for Lower Section (mm/s)	3.29	3.98	4.89	3.96	4.05	3.31	3.37	3.45
Average Ablation Rate for Middle Section (mm/s)	12.34	9.36	12.63	9.97	12.97	11.80	16.96	12.76
Average Ablation Rate for Upper Section (mm/s)	23.44	14.12	12.72	12.79	20.45	17.94	35.23	18.13

From Table II, it is seen that the inventive examples containing liquid EPDM as their exclusive EPDM ingredient (i.e., no solid EPDM) exhibited comparative, and
 5 in some instances improved ablative properties to Comparative Example A (containing less than half liquid EPDM based on the total weight of EPDM ingredients) and Comparative Example B (containing no liquid EPDM).

Examples 7 to 9 were prepared in accordance with a second embodiment of this invention, by kneading the insulation in a Buss Kneader®. The ingredients of the
 10 insulation compositions of examples 7-9 are set forth below in Table III. The ablative properties of examples 7-9, a comparison of these properties to that of inventive example 4, are set forth below in Table IV.

TABLE III
 (all units in parts by weight)

Ingredient	EXAMPLE		
	7	8	9
DSM Keltan 1446A [solid EPDM]	100	100	100
Fortafil 243 [carbon fibers]	40.52	44.45	55.55
Cure/Filler	14.86	14.88	14.88
Fire Retardant	47.02	62.72	51.62
Total Parts by Weight	202.40	222.05	222.05

TABLE IV

	EXAMPLE			
	7	8	9	4
Average Ablation Rate for Lower Section (mm/s)	3.68	2.85	3.44	3.35
Average Ablation Rate for Middle Section (mm/s)	9.41	8.97	9.79	9.40
Average Ablation Rate for Upper Section (mm/s)	15.42	14.75	13.61	11.64

As shown by Table IV, the insulation prepared in a Buss Kneader® without any liquid EPDM exhibited comparable erosion resistance to Example 4, which was
5 prepared in a sigma mixer with liquid EPDM.

The tests were performed in a char motor, such as the one illustrated in FIG. 7. Char motors are constructed to evaluate the ablative properties of solid rocket motor case insulation materials. A char motor includes a propellant beaker 70 to provide the combustion gases, evaluation chambers to hold the test materials, and a constricting
10 nozzle to produce the required pressure. The evaluation chamber is divided into three sections. The first one is a “low velocity” cylindrical region 72 about eight inches long and eight inches in diameter (approximately the same diameter as the propellant beaker). A short conical transition chamber 74 constricts the gas flow into a diameter of about 2 inches and vents the propellant gases into a 22 inch long conical test
15 chamber. This test chamber is divided into the “mid-velocity” region 76 and “high-velocity” region 78.

Samples of insulation material to be evaluated are molded, cured, and bonded with epoxy into each of the test chambers. Prior to assembly, the cured length is determined and the thickness of each evaluation material is measured at selected
20 intervals, nominally one inch apart. Each sample is also weighed. The samples are then assembled into the low-velocity section, the mid-velocity section, and the high-velocity section. After firing, the motor is disassembled, each sample is measured again. The ablation rate is determined by subtracting the post-fired thickness of the insulation of virgin insulation (i.e., after the char had been removed) at a given point

from the pre-fired thickness and dividing the result by the burn time of the motor. For these tests, more than one section of material was measured, and the average of all of the sections are reported above.

The char motors were fired with RSRM TP-H1148 (polybutadieneacrylic acid acrylonitrile (PBAN-based)) propellant. For examples 1-3 and comparative example A set forth in Tables I and II, the motor was fired for 12.10 seconds at an average pressure of 880 psi. For examples 4-6 set forth in Tables I and II, the motor was fired for 11.56 seconds at an average pressure of 825 psi. Comparative example B was fired for 11.89 seconds at an average pressure of 885 psi. For examples 7-9 and 4 set forth in Tables III and IV, the motor was fired for 12.46 seconds at an average pressure of 916.07 psi.

The foregoing detailed description of the invention has been provided for the purpose of explaining the principles of the invention and its practical application, thereby enabling others skilled in the art to understand the invention for various embodiments and with various modifications as are suited to the particular use contemplated. This description is not intended to be exhaustive or to limit the invention to the precise embodiments disclosed. Modifications and equivalents will be apparent to practitioners skilled in this art and are encompassed within the spirit and scope of the appended claims.

WHAT IS CLAIMED IS:

1. A method of insulating a case of a rocket motor loaded with a solid propellant, said method comprising:
 - 5 preparing insulation from a composition comprising a crosslinkable liquid polymer and carbon fibers, the composition including the crosslinkable liquid polymer in a sufficient concentration to permit the carbon fibers to be dispersed into the composition by mixing under substantially solvent-free conditions; and
 - curing the composition to form the insulation and insulating the case of the
 - 10 rocket motor with the insulation.
2. A method of insulating a case of a rocket motor loaded with a solid propellant, said method comprising:
 - 15 preparing insulation from a composition comprising a crosslinkable EPDM terpolymer and carbon fibers, the crosslinkable EPDM terpolymer comprising a crosslinkable liquid EPDM terpolymer in a sufficient concentration to permit the carbon fibers to be dispersed into the composition by mixing under substantially solvent-free conditions; and
 - curing the composition to form the insulation and insulating the case of the
 - 20 rocket motor with the insulation.
3. The method of claim 2, wherein said insulating comprises applying the insulation to an interior surface of the case and interposing the insulation between the interior surface and the solid propellant.

4. The method of claim 3, wherein 100 weight percent of the crosslinkable EPDM terpolymer in the composition consists of the crosslinkable liquid EPDM terpolymer.

5 5. The method of claim 3, wherein at least about 95 weight percent of the crosslinkable EPDM terpolymer in the composition consists of the crosslinkable liquid EPDM terpolymer.

10 6. The method of claim 3, wherein at least about 90 weight percent of the crosslinkable EPDM terpolymer in the composition consists of the crosslinkable liquid EPDM terpolymer.

7. The method of claim 3, wherein said preparing of the insulation comprises mixing the composition in a vertical-blade mixer or sigma-blade mixer.

15

8. The method of claim 3, wherein at least about 50 weight percent of the crosslinkable EPDM terpolymer in the composition consists of the crosslinkable liquid EPDM terpolymer, and further wherein said preparing of the insulation comprises mixing the composition in a sigma-blade mixer.

20

9. The method of claim 2, wherein the composition comprises dry ingredients, and wherein said preparing of the insulation comprises mixing the composition under conditions in which the composition comprises not more than

about 5 weight percent of volatile solvent based on the dry ingredients in the composition.

10. The method of claim 2, wherein said preparing of the insulation
5 comprises mixing the composition in the absence of any volatile solvent.

11. A method of insulating a case of a rocket motor loaded with a solid propellant, said method comprising:

providing a kneader comprising at least one housing module provided with a
10 chamber, a plurality of kneader pins extending radially into the chamber from the housing module, and a screw which is rotatable about a longitudinal axis thereof and movable along the longitudinal axis in a reciprocating motion, the screw having a peripheral surface including a plurality of screw flights arranged to collectively define a discontinuous screw thread extending along the longitudinal axis of the screw;

15 preparing insulation from a composition comprising at least one crosslinkable polymer and carbon fibers, said preparing of the insulation comprising introducing the composition into the kneader, and rotating the screw along the longitudinal axis thereof while superimposing the longitudinal reciprocating motion to the screw; and

curing the composition to form the insulation and insulating the case of the
20 rocket motor with the insulation.

12. A method of insulating a case of a rocket motor loaded with a solid propellant, said method comprising:

providing a kneader comprising at least one housing module provided with a
25 chamber, a plurality of kneader pins extending radially into the chamber from the housing module, and a screw which is rotatable about a longitudinal axis thereof and

movable along the longitudinal axis in a reciprocating motion, the screw having a peripheral surface including a plurality of screw flights arranged to collectively define a discontinuous screw thread extending along the longitudinal axis of the screw;

5 preparing insulation from a composition comprising at least one crosslinkable EPDM terpolymer and carbon fibers, said preparing of the insulation comprising introducing the composition into the kneader, and rotating the screw along the longitudinal axis thereof while superimposing the longitudinal reciprocating motion to the screw; and

10 curing the composition to form the insulation and insulating the case of the rocket motor with the insulation.

15

13. The method of claim 12, wherein said preparing of the insulation comprises mixing the composition in the kneader under substantially solvent-free conditions.

14. The method of claim 12, wherein said insulating of the case comprises applying the insulation to an interior surface of the case and interposing the insulation between the interior surface and the solid propellant.

20 15. The method of claim 12, wherein 100 weight percent of the crosslinkable EPDM terpolymer consists of crosslinkable liquid EPDM terpolymer.

25 16. The method of claim 12, wherein at least about 90 weight percent of the crosslinkable EPDM terpolymer consists of crosslinkable liquid EPDM terpolymer.

17. The method of claim 12, wherein at least about 50 weight percent of the crosslinkable EPDM terpolymer consists of crosslinkable liquid EPDM terpolymer.

5

18. The method of claim 12, wherein the composition is free of liquid EPDM.

19. The method of claim 12, wherein the composition comprises dry ingredients, and wherein said preparing of the insulation comprises mixing the composition under conditions in which the composition comprises not more than about 5 weight percent of volatile solvent based on dry ingredients in the composition.

20. The method of claim 12, wherein said preparing of the insulation comprises mixing the composition in the kneader in the absence of any volatile solvent.

1/5

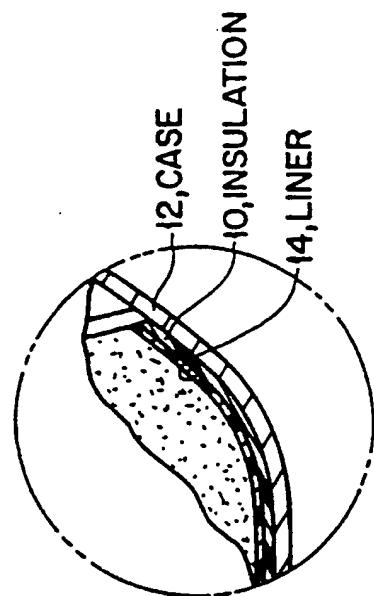


FIG. 1B

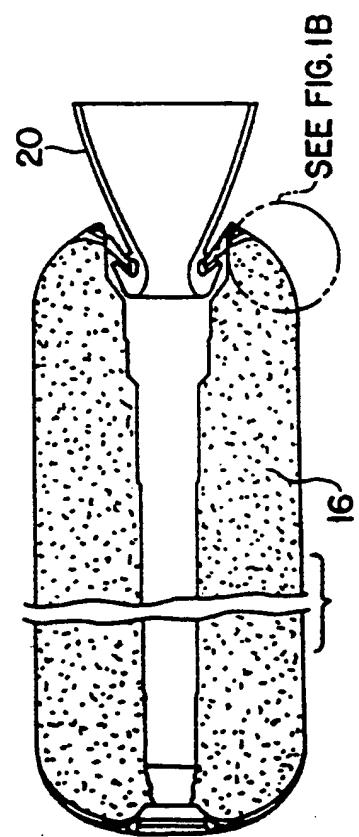
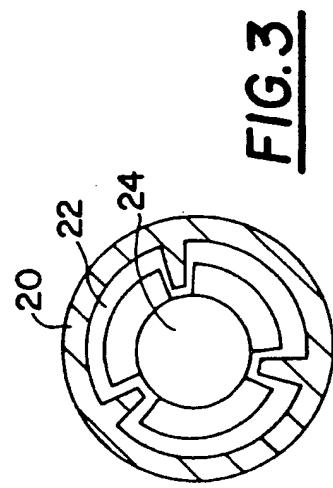
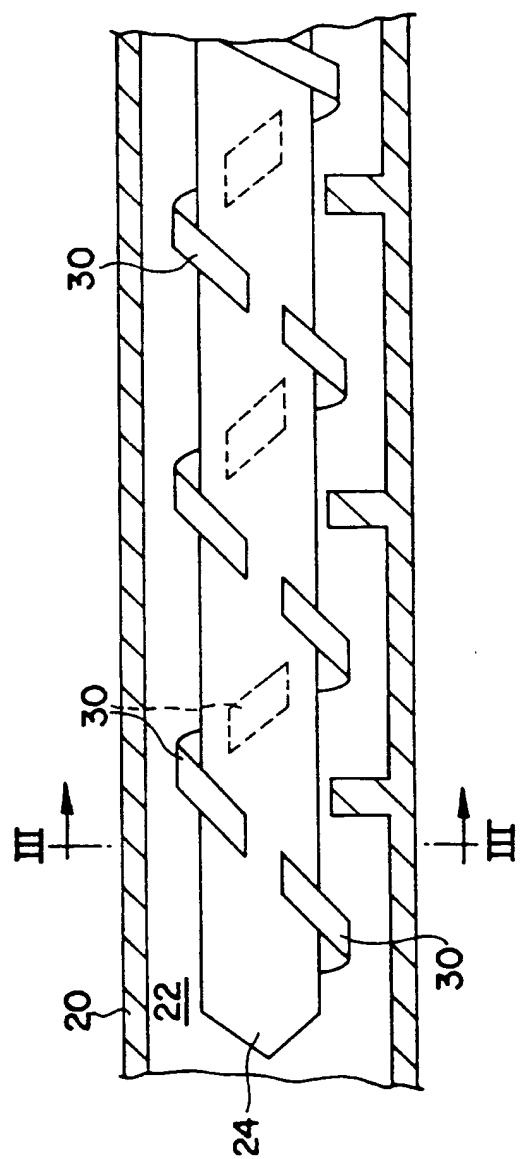


FIG. 1A

2/5



3/5

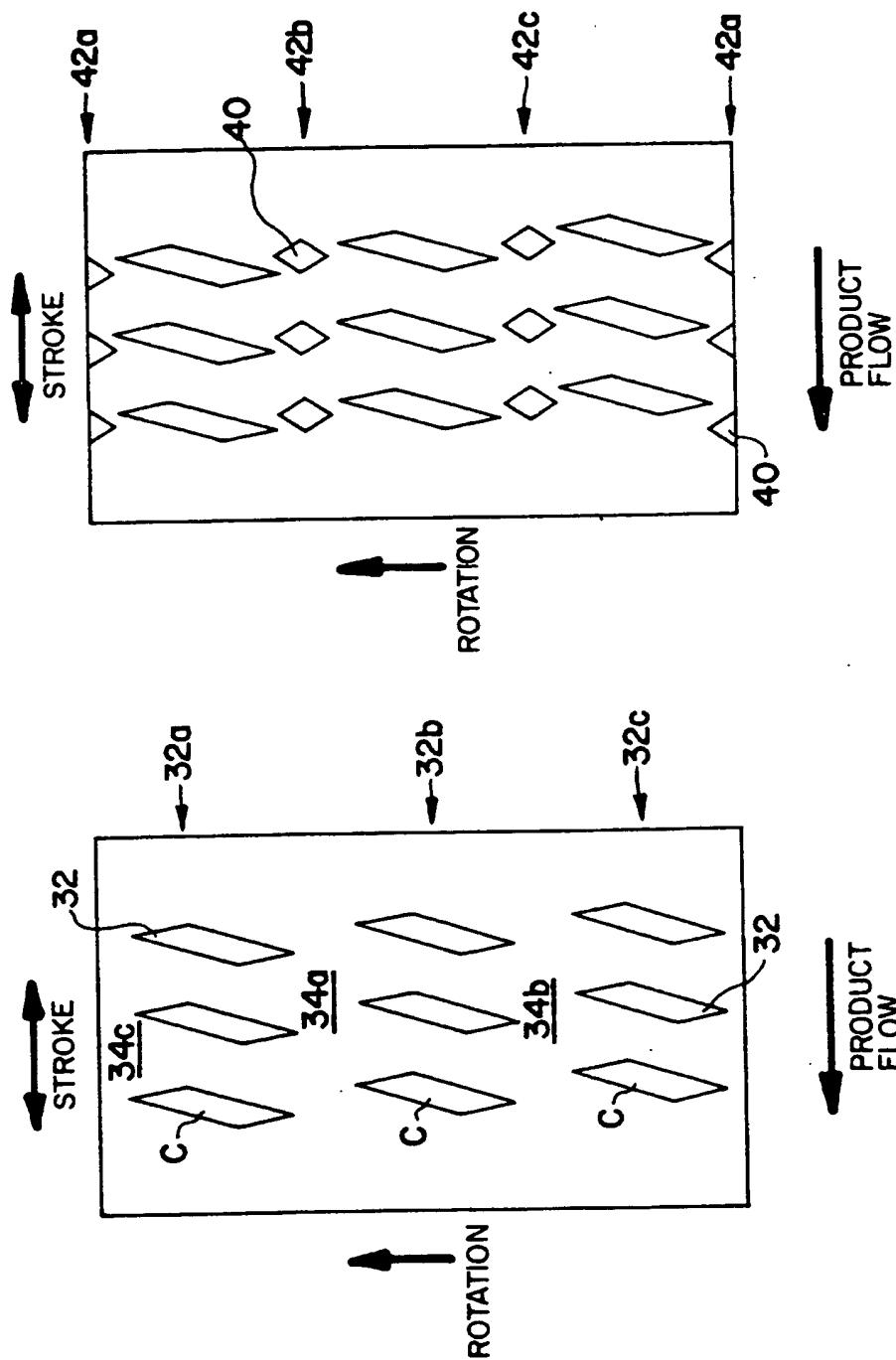


FIG. 5

FIG. 4

4/5

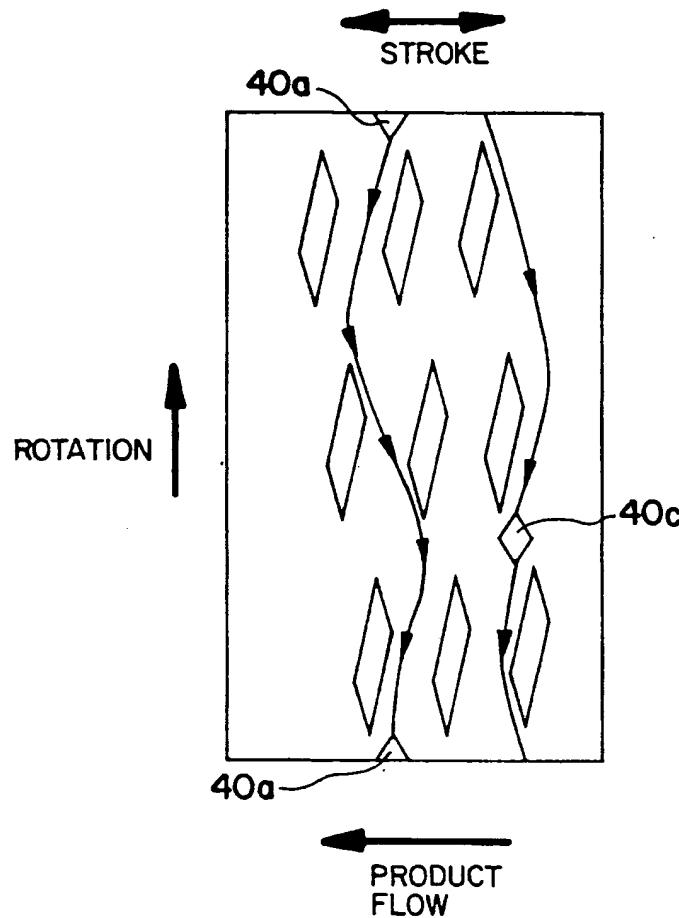


FIG. 6

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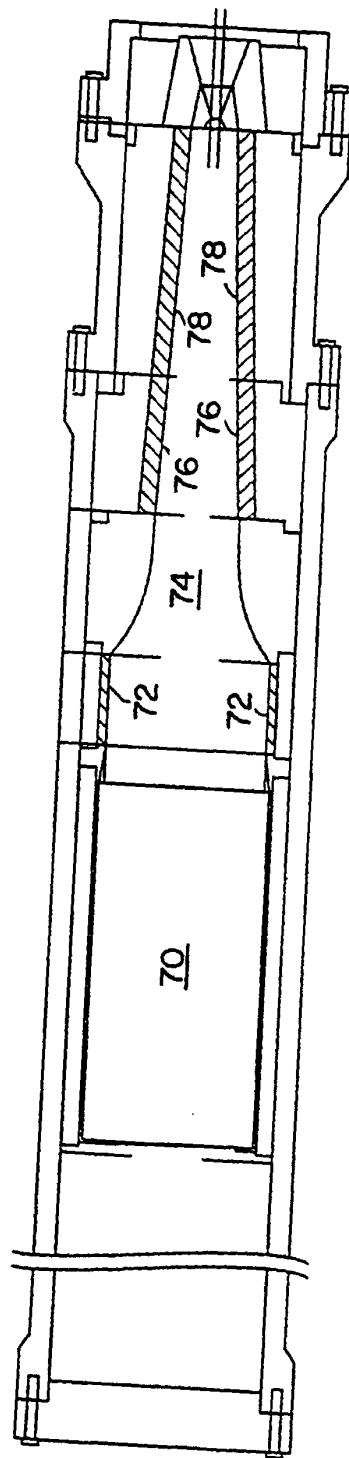


FIG. 7

INTERNATIONAL SEARCH REPORT

Intern: al Application No

PCT/US 00/35045

A. CLASSIFICATION OF SUBJECT MATTER
 IPC 7 C08F210/16 F02K9/34

According to International Patent Classification (IPC) or to both national classification and IPC

B. FIELDS SEARCHED

Minimum documentation searched (classification system followed by classification symbols)
 IPC 7 C08F F02K

Documentation searched other than minimum documentation to the extent that such documents are included in the fields searched

Electronic data base consulted during the international search (name of data base and, where practical, search terms used)

EPO-Internal, WPI Data, PAJ

C. DOCUMENTS CONSIDERED TO BE RELEVANT

Category °	Citation of document, with indication, where appropriate, of the relevant passages	Relevant to claim No.
A	FR 2 554 114 A (HERCULES INC) 3 May 1985 (1985-05-03) the whole document ---	1-20
A	US 5 352 312 A (GUILLOT DAVID G) 4 October 1994 (1994-10-04) the whole document ---	1-20
A	US 5 767 221 A (SMALLEY JR ROBERT B ET AL) 16 June 1998 (1998-06-16) cited in the application the whole document ---	1-20

Further documents are listed in the continuation of box C.

Patent family members are listed in annex.

* Special categories of cited documents :

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Date of the actual completion of the international search

5 April 2001

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Name and mailing address of the ISA

European Patent Office, P.B. 5818 Patentlaan 2
 NL - 2280 HV Rijswijk
 Tel. (+31-70) 340-2040, Tx. 31 651 epo nl.
 Fax: (+31-70) 340-3016

Authorized officer

Van Golde, L

INTERNATIONAL SEARCH REPORT

Information on patent family members

Intern. Appl. No.

PCT/US 00/35045

Patent document cited in search report	Publication date	Patent family member(s)	Publication date
FR 2554114 A	03-05-1985	NONE	
US 5352312 A	04-10-1994	NONE	
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